

WHAT IS CLAIMED IS:

1. A blade structure in a gas turbine, comprising:
stationary blades arrayed in a circle on a casing;
moving blades arrayed in a circle on a rotor, wherein
5 a clearance is provided between chips of the moving blades
and the casing, wherein
a front-edge including angle at a chip portion of the
stationary blade that is the stationary blade at the rear
stage of the moving blade having the chip clearance is larger
10 than a front-edge including angle at other portions than
the chip portion of the stationary blade.
2. A blade structure in a gas turbine, comprising:
stationary blades arrayed in a circle on a casing;
15 moving blades arrayed in a circle on a rotor, wherein
a clearance is provided between chips of the moving blades
and the casing, wherein
an entrance metal angle at a chip portion of the
stationary blade that is the stationary blade at the rear
20 stage of the moving blade having the chip clearance is smaller
than an entrance metal angle at other portions than the chip
portion of the stationary blade.

3. A blade structure in a gas turbine, comprising:
stationary blades arrayed in a circle on a casing;
moving blades arrayed in a circle on a rotor, wherein
a clearance is provided between chips of the moving blades
5 and the casing, wherein
a front-edge including angle at a chip portion of the
stationary blade that is the stationary blade at the rear
stage of the moving blade having the chip clearance is larger
than a front-edge including angle at other portions than
10 the chip portion of the stationary blade, and also an entrance
metal angle at a chip portion of the stationary blade is
smaller than an entrance metal angle at other portions than
the chip portion of the stationary blade.
- 15 4. A blade structure in a gas turbine, comprising:
stationary blades arrayed in a circle on a casing;
moving blades arrayed in a circle on a rotor, wherein
seal-air flows from the rotor side at the upstream of the
moving blades, wherein
20 a front-edge including angle at a hub portion of the
stationary blade is larger than a front-edge including angle
at other portions than the hub portion of the moving blade.

5. A blade structure in a gas turbine, comprising:
stationary blades arrayed in a circle on a casing;
moving blades arrayed in a circle on a rotor, wherein
seal-air flows from the rotor side at the upstream of the
moving blades, wherein
an entrance metal angle at a hub portion of the
stationary blade is smaller than an entrance metal angle
at other portions than the hub portion of the moving blade.
6. A blade structure in a gas turbine , comprising:
stationary blades arrayed in a circle on a casing;
moving blades arrayed in a circle on a rotor, wherein
seal-air flows from the rotor side at the upstream of the
moving blades, wherein
a front-edge including angle at a hub portion of the
stationary blade is larger than a front-edge including angle
at other portions than the hub portion of the moving blade,
and also an entrance metal angle at a hub portion of the
stationary blade is smaller than an entrance metal angle
at other portions than the hub portion of the moving blade.
7. A blade structure in a gas turbine , comprising:
stationary blades arrayed in a circle on a casing;
moving blades arrayed in a circle on a rotor, wherein
a clearance is provided between chips of the moving blades

and the casing, wherein

a chord length at a chip portion of the moving blade having the chip clearance is larger than a minimum chord length at other portions than the chip portion of the moving
5 blade.

8. The blade structure in a gas turbine according to claim 7, wherein the chip portion of the stationary blade is provided with an escape section for avoiding an interference
10 with the chip portion of the moving blade.

9. The blade structure in a gas turbine according to claim 8, wherein the escape section of the chip portion of the stationary blade is arranged such that an entrance metal
15 angle at a chip portion of the stationary blade is smaller than an entrance metal angle at other portions than the chip portion of the stationary blade, and that the entrance metal angle at the chip portion of the stationary blade is directed toward the rear surface side of the stationary blade.

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